Patent Claims

1. A gas turbine vane, in particular a vane of an aircraft engine, comprising a blade (11) and a vane foot (12), the blade (11) being delimited by a flow inlet edge or front edge (13), a flow outlet edge or rear edge (14), and a blade surface (15) extending between the front edge (13) and the rear edge (15) and forming a suction side (16) and a pressure side (17),
characterized in that the suction side (17) of the blade (11) has at least one micro-profiled or micro-structured area (18; 20, 21, 22) for optimizing the flow around the blade (11).

2. The gas turbine vane as recited in Claim 1,
characterized in that the or each micro-profiled or micro-structured area (18; 20, 21, 22) is assigned to a section of the suction side (17) of the blade (11) in which a flow deceleration takes place.

3. The gas turbine vane as recited in Claim 1 or 2,
characterized in that the or each micro-profiled or micro-structured area (18; 20, 21, 22) is assigned to a section of the suction side (16) of the blade (11) which extends over between 30% and 70% of the profile depth of the blade (11).

4. The gas turbine vane as recited in Claim 3,
characterized in that the or each micro-profiled or micro-structured area (18; 20, 21, 22) is assigned to a section of the suction side (16) of the blade (11) which extends over between 30% and 50% of the profile depth of the blade (11).

5. The gas turbine vane as recited in one or more of Claims 1 through 4,
characterized in that the or each micro-profiled or micro-structured area (18; 20, 21, 22) has a shark skin-like profile or structure.

6. The gas turbine vane as recited in one or more of Claims 1 through 5,
characterized in that a micro-profiled or micro-structured area (20), situated on the side of the

vane foot, is structured in such a way that the blade (11) is strengthened in this area and/or that compressive stresses are induced.

7. The gas turbine vane as recited in one or more of Claims 1 through 6, characterized in that it is designed as a rotary blade of a compressor.